

CLAIMS

1. A combustor for a gas turbine, comprising:
 a first burner injecting a fuel and an air
into a combustion chamber; and
 a second burner generating a circulation jet
flow of the fuel and the air at a position
corresponding to a leading end portion of a flame
generated by the first burner.
2. A combustor for a gas turbine, comprising:
 a first burner injecting a fuel and an air
into a combustion chamber; and
 a second burner spraying the fuel and the air
so as to intersect a downstream side of a flame
generated by the first burner.
3. A combustor for a gas turbine, comprising:
 a first burner injecting a fuel and an air
into a combustion chamber; and
 a second burner guiding the fuel and the air
so as to intersect a distributing direction of a flame
generated by the first burner.
4. A combustor for a gas turbine as claimed in
claim 1, 2 or 3, wherein the second burner is provided
so as to pass through a peripheral wall forming the
combustion chamber.
5. A combustor for a gas turbine as claimed in
claim 1, 2 or 3, wherein the second burner is
constituted by a plurality of burners, and these
plurality of burners are arranged in such a manner that

the fuel and the air come into collision with each other near a center portion of the combustion chamber.

6. A combustor for a gas turbine as claimed in claim 1, 2 or 3, wherein the second burner is provided with a fuel injection nozzle near a center portion of the combustion chamber, such that the fuel is positioned in an outer side of a spray flow of the air.

7. A combustor for a gas turbine as claimed in claim 1, 2 or 3, wherein the second burner is provided with a guide tube guiding the fuel and the air to a center portion of the combustion chamber, in a peripheral wall forming the combustion chamber, and the guide tube protrudes into the combustion chamber.

8. A combustor for a gas turbine, comprising:
a first burner injecting a fuel and an air into a combustion chamber;

a second burner generating a circulation jet flow of the fuel and the air at a position corresponding to a leading end portion of a frame generated by the first burner; and

a third burner generating a circulation jet flow of an air-fuel mixture near a terminal end portion of a reaction region within the combustion chamber.

9. A combustor for a gas turbine comprising:

a pilot burner securing a combustion stability in an upstream side of a combustion chamber; and

a lean air-fuel mixture guiding means

generating a circulation jet flow of a lean air-fuel mixture at a leading end portion of a flame generated by the pilot burner.